



# Detecting Exomoons in Free-floating-planet Events from Space-based Microlensing Surveys

Haozhu Fu<sup>1,2</sup> and Subo Dong<sup>1,2,3</sup>

<sup>1</sup> Department of Astronomy, Peking University, Beijing 100871, China; [dongsubo@pku.edu.cn](mailto:dongsubo@pku.edu.cn)

<sup>2</sup> Kavli Institute of Astronomy and Astrophysics, Peking University, Beijing 100871, China

<sup>3</sup> National Astronomical Observatories, Chinese Academy of Sciences, Beijing 100101, China

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## Abstract

When a planet is ejected from its star-planet system due to dynamical interactions, its satellite may remain gravitationally bound to the planet. The Chinese Space Station Telescope (CSST) will be capable of detecting a large number of low-mass free-floating planet events (FFPs) from a bulge microlensing survey. We assess the feasibility of detecting satellites (a.k.a., exomoons) orbiting FFPs by simulating CSST light curves and calculating the detection efficiency as a function of satellite-to-planet mass ratios ( $q$ ) and projected separations ( $s$ ) in units of the Einstein radius. For a Neptune-class FFP in the Galactic disk with a Sun-like star as the microlensed source, CSST can detect Earth-mass satellites over a decade of separations ( $\sim 0.01$ – $0.1$  au) and has sensitivity down to Moon-mass satellites ( $q \sim 10^{-3}$ ) at  $s \sim 1$ . CSST also has some sensitivity to detect Moon-mass satellites at  $s \sim 2$  ( $\sim 0.02$  au) orbiting an Earth-mass FFP in the disk. CSST has substantially reduced sensitivity for detecting satellites when the source star is an M dwarf, compared to a Sun-like source. We also calculate the satellite detection efficiency for the dedicated microlensing survey of the Roman Space Telescope (Roman), which demonstrates greater sensitivity than CSST, particularly for M-dwarf sources. Notably, some of the Neptune-Earth systems detectable by CSST and Roman may exhibit significant tidal heating.

**Key words:** planets and satellites: general – planets and satellites: detection – gravitational lensing: micro

## 1. Introduction

A planet gravitationally unbound to any host stars can be detected in a short-duration ( $< 1$  day) microlensing event. In the event, the planet acts as a “lens” bending the light from a background star (“source”) in the observer’s sightline. More than a dozen such free-floating-planet candidates (hereafter FFPs) have been found (for recent reviews, see Zhu & Dong 2021; Mroz & Poleski 2023). An FFP event can also be due to a planetary lens on a wide-separation ( $\gtrsim 20$  au) orbit from a host star, which can be distinguished from an unbound planet by follow-up adaptive-optics observations with instruments such as ELT-MICADO (Sturm et al. 2024) a few years after the event. Statistical analyses on three independent samples from OGLE, KMTNet, and MOA surveys suggest that low-mass (Earth-mass to Neptune-class) FFPs are likely common in the Galaxy, possibly a few times more numerous than stars (Mróz et al. 2017; Gould et al. 2022; Sumi et al. 2023).

A class of mechanisms to generate unbound planets is dynamical ejection from planetary systems via planet–planet scatterings (e.g., Rasio & Ford 1996; Weidenschilling & Marzari 1996; Chatterjee et al. 2008; Jurić & Tremaine 2008). The ejection likely occurs from a perturber in the outer planetary system ( $a > 1$  au) with the Safronov number larger than unity, that is, the escape velocity at the perturber’s surface

being greater than the orbital escape velocity. Previous works (Debes & Sigurdsson 2007; Hong et al. 2018; Rabago & Steffen 2019) suggest that, when a planet is ejected, its satellite has a reasonable probability of remaining gravitationally bound to the planet. Free-floating planet-satellite systems are also studied for habitability (Reynolds et al. 1987; Scharf 2006) in that they might be able to maintain the existence of liquid water due to tidal heating (Reynolds et al. 1987; Scharf 2006).

To date, there have been a handful of satellite candidates (i.e., exomoons) around bound planets identified by microlensing or transits but with no definitive detection (see, e.g., Bennett et al. 2014; Teachey & Kipping 2018; Kreidberg et al. 2019). Because the finite-source effects (Gould 1994; Nemiroff & Wickramasinghe 1994; Witt & Mao 1994) could significantly reduce the amplitude of a satellite’s signal, a space-based microlensing survey offers the most promising opportunity for detecting satellites via microlensing (Bennett & Rhie 2002; Han & Han 2002; Liebig & Wambsganss 2010).

In this paper, we focus on studying the prospects of detecting satellites orbiting FFPs in a bulge microlensing survey using the Chinese Space Station Telescope (CSST)<sup>4</sup> which is a planned 2 m space telescope in a low-Earth orbit ( $\approx 1.5$  hr period). CSST

<sup>4</sup> <https://nadc.china-vo.org/csst-bp/article/20230707113736>.

will be equipped with a wide-field ( $\sim 1.1 \text{ deg}^2$ ) survey camera designed to take diffraction-limited images at optical wavelengths. We also conduct simulations for the Nancy Grace Roman Space Telescope (a.k.a., WFIRST; hereafter Roman), whose Galactic Bulge Time Domain Survey aims to discover bound planets and FFPs with microlensing (Penny et al. 2019; Johnson et al. 2020; Yee & Gould 2023). Recently, Sajadian & Sangtarash (2023) have studied Roman’s detection efficiency of free-floating planet-satellite systems, and we compare our results with theirs.

## 2. Simulating Microlensing Observations

The Einstein radius sets the basic scale in microlensing, and its angular size is,

$$\theta_E = \sqrt{\kappa M_L \pi_{\text{rel}}}, \quad (1)$$

where

$$\pi_{\text{rel}} = \text{au} \left( \frac{1}{D_L} - \frac{1}{D_S} \right), \quad (2)$$

is the relative trigonometric parallax between the lens at distance  $D_L$  and the source at  $D_S$ ,  $M_L$  is the lens mass and  $\kappa = 4\pi G/(c^2 \text{au}) = 8.144 \text{ mas } M_\odot^{-1}$  is a constant. The physical Einstein radius is  $R_E = D_L \theta_E$ .

The ultra-short microlensing events with measured  $\theta_E$  provide compelling evidence for the existence of FFPs (e.g., Mróz et al. 2018). They collectively have  $\theta_E \lesssim 9 \mu\text{as}$ , below an empirical gap in the  $\theta_E$  distribution between  $\sim 9 \mu\text{as}$  and  $\sim 25 \mu\text{as}$  (the “Einstein desert”), suggesting that they belong to a planetary population separated from brown dwarfs and low-mass stars (Ryu et al. 2021; Gould et al. 2022). The  $\theta_E$  measurements are made for finite-source-point-lens (FSPL) events, during which the lens transits the source with angular radius  $\theta_*$  being larger than or comparable to  $\theta_E$ . From fitting the light curve of an FSPL event, the scaled source size  $\rho = \theta_*/\theta_E$  can be directly extracted, and then  $\theta_E$  is estimated by using  $\theta_*$  measured via the source’s color and apparent magnitude (Yoo et al. 2004b).

In an FSPL event, the magnification  $A(t)$  is a function of  $(t_0, u_0, t_E, \rho)$ , where  $t_E = \theta_E/\mu_{\text{rel}}$  is the time taken by the source at a relative proper motion  $\mu_{\text{rel}}$  with respect to the lens to cross  $\theta_E$ ,  $u_0$  is the impact parameter and  $t_0$  is the time of the peak. Introducing a satellite into the FFP lens system requires three additional binary-lens parameters  $(s, q, \alpha)$ , where  $q = m_{\text{satellite}}/m_{\text{planet}}$  is the satellite-planet mass ratio,  $s$  is the angular distance between the planet and the satellite scaled by  $\theta_E$ , and  $\alpha$  is the angle between the trajectory of the source and the satellite-planet vector.

We adopt the approach of Yan & Zhu (2022) to simulate the surveys. We estimate CSST photometric uncertainties using the Exposure Time Calculator<sup>5</sup> for an exposure time of 60 s in

**Table 1**  
Adopted Source Properties

| Source Type | $\theta^*/\mu\text{as}$ | $i$ mag | W149 mag |
|-------------|-------------------------|---------|----------|
| G2V         | 0.56                    | 20      | 19.6     |
| M0V         | 0.28                    | 23      | 21.7     |

**Table 2**  
Adopted Planet Masses

| Planet Type   | Mass/ $M_\odot$      | Mass/ $M_\oplus$ | $\log(\text{Mass}/m_{\text{Moon}})$ |
|---------------|----------------------|------------------|-------------------------------------|
| Earth-mass    | $3 \times 10^{-6}$   | 1                | 1.91                                |
| Neptune-class | $4.8 \times 10^{-5}$ | 16               | 3.11                                |

the  $i$ -band and a systematic noise floor of 0.001 mag. The duty cycle is 40%, and for each orbit, there are eight observations (i.e.,  $\sim 5$  minutes cadence). We adopt a 15 minutes cadence for Roman and assume the photometric performance in W149 according to Penny et al. (2019).

We simulate microlensing events with a set of representative parameters for the source and lens. Following Yan & Zhu (2022), we consider two types of sources: early M-dwarfs (M0V) and Sun-like dwarfs (G2V) in the Galactic bulge ( $D_S = 8.2 \text{ kpc}$ ), adopting the source sizes of  $\theta_* = 0.28 \mu\text{as}$  and  $0.56 \mu\text{as}$ , respectively. We first estimate the Johnson-Cousin  $I$ - and  $H$ -band magnitudes based on Pecaut & Mamajek (2013), and we apply extinction corrections by adopting  $A_I = 1.5$  and  $E(I - H) = 1$  (Gonzalez et al. 2012; Nataf et al. 2013). Then we convert the Johnson-Cousin magnitudes in the Vega system to the AB system using results from Blanton & Roweis (2007), deriving that the M0V (G2V) source has  $m_i \approx 23$  (20) mag and  $m_{W149} \approx 21.7$  (19.6) mag, corresponding to uncertainties in baseline magnitudes of  $\approx 0.05$  (0.01) mag and  $\approx 0.014$  (0.005) mag, respectively. The adopted source properties are listed in Table 1.

The lens is placed in either the bulge or the disk. For a bulge lens in our simulations, the relative parallax is  $\pi_{\text{rel}} = 0.02 \text{ mas}$  ( $D_L \approx 7 \text{ kpc}$ ) with a relative proper motion of  $\mu_{\text{rel}} = 4 \text{ mas yr}^{-1}$ . For a disk lens,  $\pi_{\text{rel}} = 0.12 \text{ mas}$  ( $D_L \approx 4 \text{ kpc}$ ) and  $\mu_{\text{rel}} = 7 \text{ mas yr}^{-1}$ . The FFP lens is assumed to have a mass of either an Earth-mass planet ( $M_L = 3 \times 10^{-6} M_\odot$ ) or a Neptune-class (i.e., using the averaged mass of Neptune and Uranus) planet ( $M_L = 4.8 \times 10^{-5} M_\odot$ ). The adopted planet properties are listed in Table 2, while the parameters adopted in our simulations are listed in Table 3.

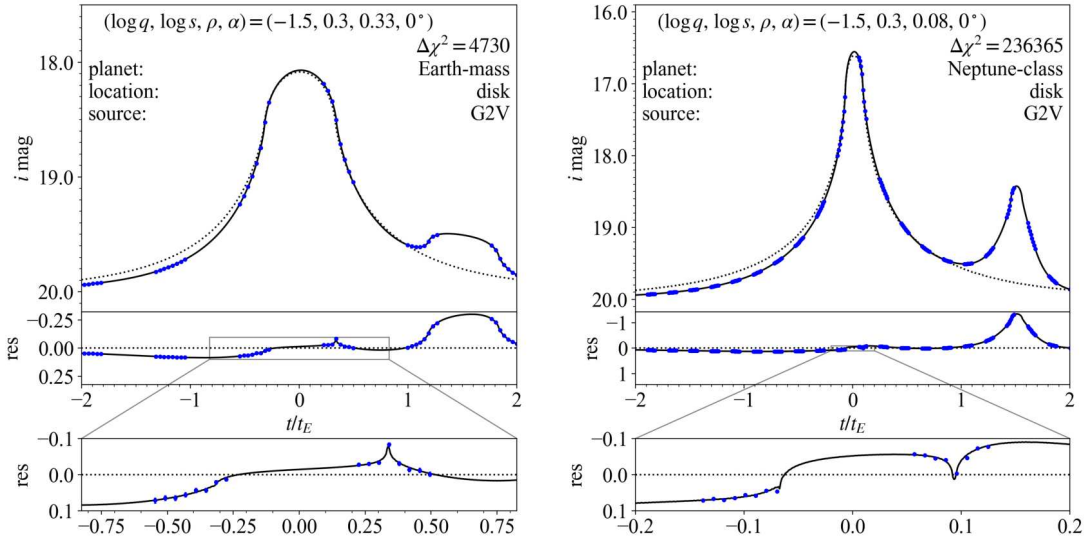
### 2.1. Detection Efficiency

We follow the commonly used method (Rhie et al. 2000; Dong et al. 2006) to estimate the detection efficiency of satellites. Mock binary-lens (i.e., planet-satellite) single-source (2L1S) light curves are simulated using the VBBinaryLensing code (Bozza et al. 2018). Then the mock data are fitted to single-

<sup>5</sup> <https://nadc.china-vo.org/csst-bp/etc-ms/etc.jsp>.

**Table 3**  
Microlensing Parameters Adopted in the Simulations

| Planet Type   | $\pi_{\text{rel}}/\text{mas}$ (Planet Location) | $\theta_E/\mu\text{as}$ | $R_E/\text{au}$ | $t_E/\text{day}$ | $\rho$ (Source Type)       |
|---------------|-------------------------------------------------|-------------------------|-----------------|------------------|----------------------------|
| Earth-mass    | 0.12 (disk)                                     | 1.70                    | 0.007           | 0.090            | 0.33 (G2V) and 0.165 (M0V) |
| Neptune-class | 0.12 (disk)                                     | 6.85                    | 0.03            | 0.357            | 0.08 (G2V) and 0.04 (M0V)  |
| Earth-mass    | 0.02 (bulge)                                    | 0.70                    | 0.005           | 0.064            | 0.80 (G2V) and 0.40 (M0V)  |
| Neptune-class | 0.02 (bulge)                                    | 2.80                    | 0.02            | 0.256            | 0.20 (G2V) and 0.10 (M0V)  |



**Figure 1.** Examples of simulated CSST 2L1S light curves of a G2V source with planet-satellite parameters of  $(\log s, \log q, \alpha) = (0.3, -1.5, 0^\circ)$ . Left: The primary lens is an Earth-mass planet in the Galactic disk ( $t_E = 0.090$  day,  $\rho = 0.33$ ). The top sub-panel shows the simulated data (blue dots), the underlying 2L1S model (black curve), and the best-fit 1L1S model (dotted curve). The middle sub-panel presents the residuals to the best-fit 1L1S model, and the zoomed-in view of the residuals around the light-curve peak is shown in the bottom sub-panel. Right: Similar to the left, but for a Neptune-class planet ( $t_E = 0.357$  day,  $\rho = 0.08$ ).

lens single-source (1L1S) models with free parameters ( $t_0$ ,  $u_0$ ,  $t_E$ ,  $\rho$ ) using Markov chain Monte Carlo (MCMC). The magnifications for 1L1S models with finite-source effects (i.e., FSPL models) are calculated with the map-making algorithm (Dong et al. 2006, 2009). A satellite is regarded as being detected if the  $\chi^2$  difference between the best-fit 1L1S model and the input 2L1S model exceeds the detection threshold ( $\Delta\chi^2 \geq 100$ ). For each set of  $(s, q)$ , we evaluate 180 uniformly distributed  $\alpha$  values within the range  $[0, 2\pi)$ . The detection efficiency for a given parameter set is defined as the fraction of  $\alpha$  values for which the satellite is detected. Then, we calculate the detection-efficiency distribution in the  $s$ - $q$  plane, covering the logarithmic ranges  $[-1.0, 1.0]$  for  $\log s$  and  $[-4.0, -1.0]$  for  $\log q$ , with step sizes of 0.033 dex and 0.1 dex, respectively. We analyze mock data with  $u_0 = 0$  by default and evaluate how non-zero  $u_0$  impacts the detection efficiency.

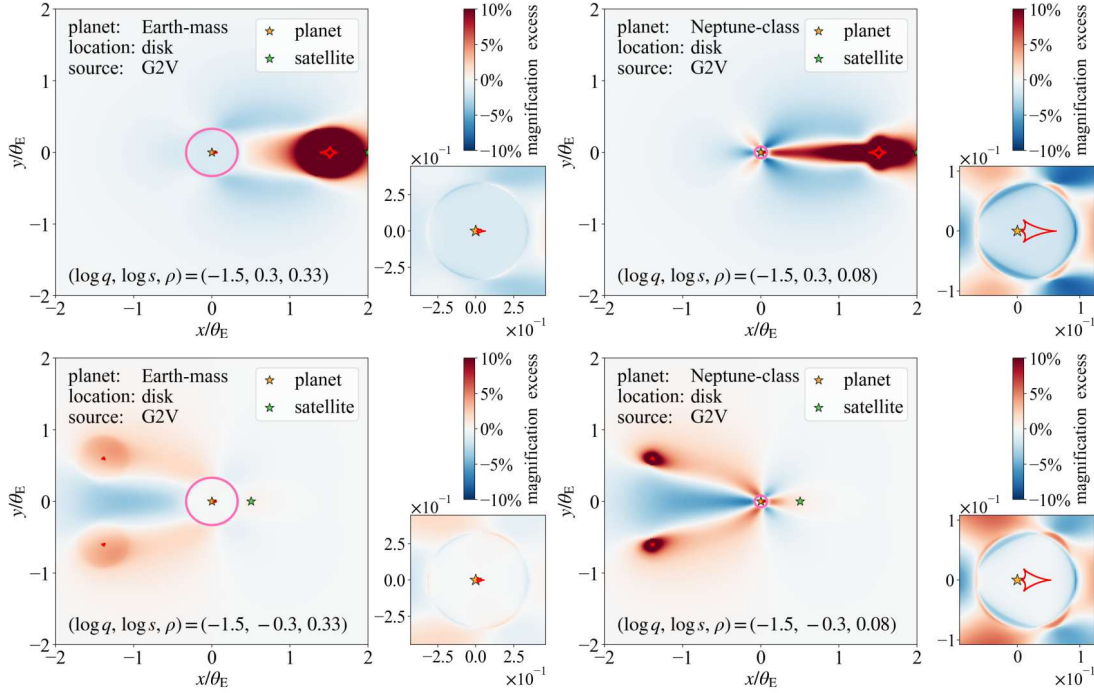
### 3. Simulation Results

In this section, we present the results of our simulations. In contrast to the ground-based surveys, which are most sensitive

to FFPs with bright giant sources, space-based surveys can probe FFPs with smaller dwarf sources thanks to their superior photometric performance (high precision and stability) and diffraction-limited resolutions. As described in Section 2, we choose two representative types of sources (G2V and M0V).

#### 3.1. G-dwarf Source

We first discuss results involving a G2V source. Figure 1 shows two examples of mock CSST light curves for detectable satellites with  $\approx 3\%$  of planetary mass ( $\log q = -1.5$ ). In both cases, the planets are in the Galactic disk, with Earth-mass (left) and Neptune-class (right), respectively. One class of satellite signals is the “central-caustics” (Griest & Safizadeh 1998) perturbations near the peak (see the bottom-left sub-panels). Central-caustic perturbations exist for essentially all position angles of the source trajectories. There are also off-peak perturbations due to caustics along the binary axis but away from the position of the primary. These caustics are called “planetary caustics” in the context of the star-planet lens system, and here we refer them as “satellite caustics” for clarity.



**Figure 2.** The magnification excess maps for planet-satellite configurations with  $\log q = -1.5$  and  $|\log s| = 0.3$ . The color coding indicates fractional magnification differences between the 2LIS and 1LIS models. The maps with wide-separation ( $\log s = 0.3$ ) and close-separation ( $\log s = -0.3$ ) cases are displayed in the upper and lower panels, respectively. The left and right panels correspond to Earth-mass and Neptune-class planets. The  $x$ - $y$  coordinates denote the source positions in units of  $\theta_E$ , with the positive  $x$ -axis aligned in the direction pointing from the planet (gold star) to the satellite (cyan star). Magenta circles indicate the source stars' sizes ( $\rho = 0.33$  for Earth-mass lenses and  $\rho = 0.08$  for Neptune-class). Caustics are depicted in red, with the central caustic located near the planet and the satellite caustics positioned farther away. Each panel includes a lower-right inset offering a zoom-in view of the central caustics.

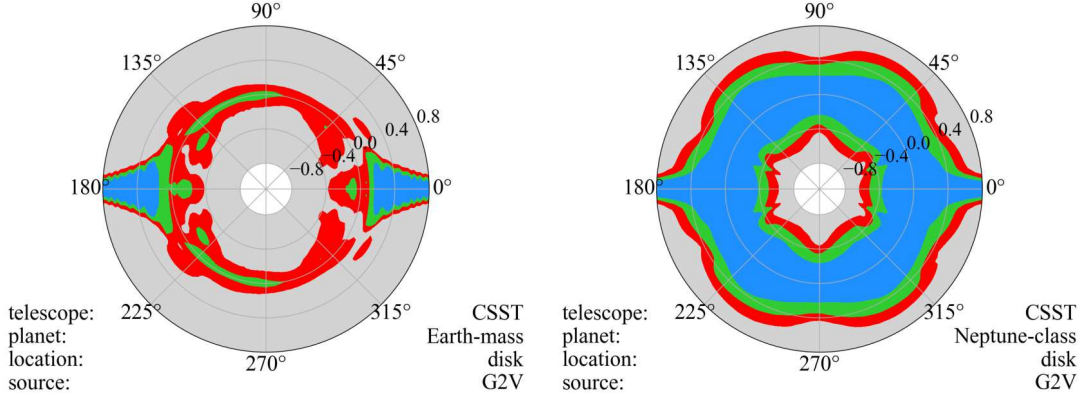
Depending on whether  $s > 1$  (wide) or  $s < 1$  (close), there are one or two satellite caustics, respectively on the same and/or opposite side of the satellite with respect to the planet. These two examples are both wide with  $s = 2.0$  ( $\log s = 0.3$ ). In contrast to central caustics, considerable perturbations are usually produced when the source trajectories cross or approach the caustic. For the examples, we pick source trajectories with  $u_0 = 0$  and  $\alpha = 0^\circ$ , which cross the satellite caustics.

We further explore the satellite perturbations using the magnification excess maps (see, e.g., Dong et al. 2009), which display the distributions of fractional differences in magnifications between the 2LIS models (shifted to the center of the magnification pattern, Yoo et al. 2004a; Dong et al. 2006) and the underlying 1LIS models. See Figure 2 for the magnification excess maps of Earth-mass and Neptune-class lenses with  $\log q = -1.5$  and  $|\log s| = 0.3$  (i.e., both wide and close). The satellite signals are concentrated in the vicinities of the caustics (red). The satellite caustics show substantially more prominent perturbations than the central caustics (highlighted in the lower-right insets of the sub-panels), and the most significant perturbations are from the satellite caustics of the wide-separation cases (upper). The signals are generally

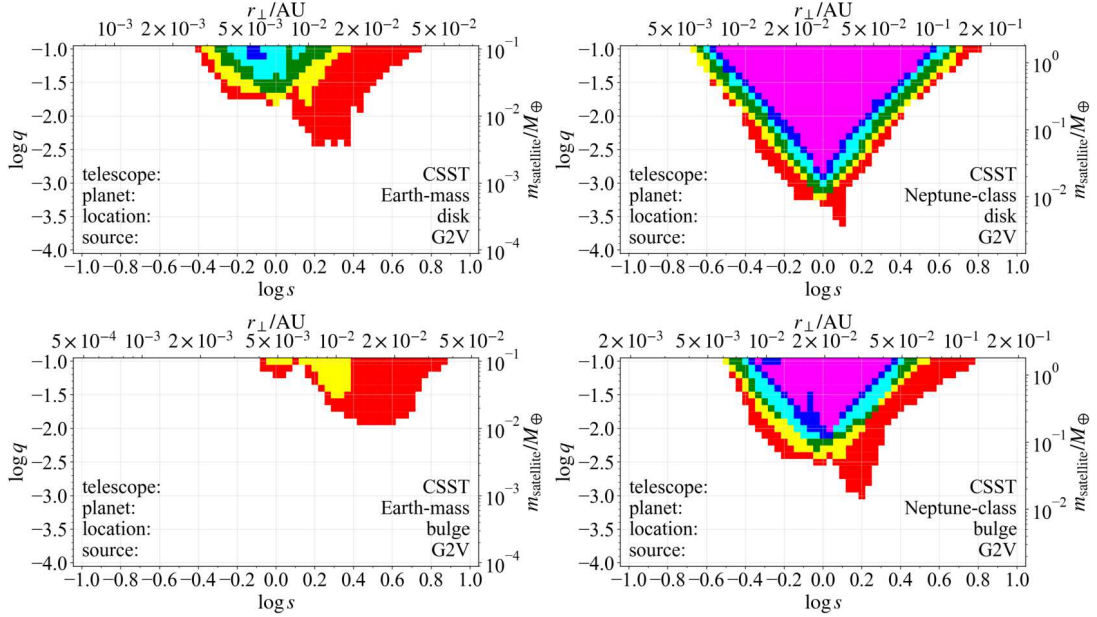
stronger for the Neptune-class planets (right) than the Earth-mass planets (left). The signals for the Earth-mass planet are more “washed out” due to the larger source size ( $\rho = 0.33$ ) than that for the Neptune ( $\rho = 0.08$ ), while the larger finite-source effects also considerably broaden the perturbation region surrounding the satellite caustics of the wide case for the Earth-mass planet.

Figure 3 presents the  $\Delta\chi^2$  distributions as functions of  $\log s$  and  $\alpha$  for mock data with  $\log q = -1.5$ . The regions with detected satellites are significantly larger for Neptune-class FFPs (right) compared to Earth-mass FFPs (left), primarily due to the smaller source sizes for the simulated Neptune-class FFP cases, as discussed above.

The top-right panel of Figure 4 displays the detection efficiency of satellites orbiting a Neptune FFP in the disk, exhibiting a triangular pattern that is symmetric with respect to  $s = 1$  ( $\log s = 0$ ). Such a pattern is characteristic of planetary detection efficiency in high-magnification events (see, e.g., Dong et al. 2006; Gould et al. 2010), arising from the close-wide ( $s$ - $s^{-1}$ ) degeneracy of the central-caustic perturbations (Griest & Safizadeh 1998; Dominik 1999; Bozza 1999; An 2005, 2021). The sensitivity zone reaches down to  $q \approx 10^{-3}$  near the Einstein



**Figure 3.**  $\Delta\chi^2$  distributions in polar coordinate ( $\log s$ ,  $\alpha$ ) at  $\log q = -1.5$  (left: Earth; right: Neptune). Regions satisfying  $\Delta\chi^2 > 100$ ,  $> 400$ , and  $> 900$  are color-coded in red, green, and blue, respectively, and the satellites are detected in these regions according to our detection threshold ( $\Delta\chi^2 > 100$ ). The satellites are undetected in the gray regions.



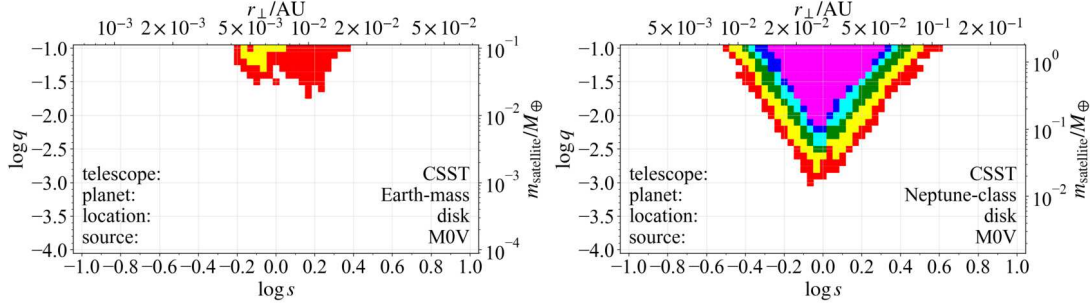
**Figure 4.** CSST satellite detection efficiency as a function of the satellite-planet mass ratio  $q$  and planet-satellite projected separation  $s$ , with a G2V source. The satellite mass  $m_{\text{satellite}}$  and physical projected separation  $r_\perp$  are also displayed. The Earth-mass (left panels) and Neptune-class (right panels) planetary lenses are located in the disk (upper panels) and bulge (lower panels), respectively. The color coding indicates efficiency levels: 10% (red), 25% (yellow), 50% (green), 75% (cyan), 90% (blue) and 100% (magenta).

radius, corresponding to a Moon-mass ( $\sim 10^{-2} M_\oplus$ ) satellite at physical projected separation  $r_\perp = s R_E \approx 3 \times 10^{-3}$  au. For Earth-mass satellites, the sensitivity covers  $\sim 1$  dex around the Einstein radius, corresponding to  $r_\perp \approx 0.01$ – $0.1$  au.

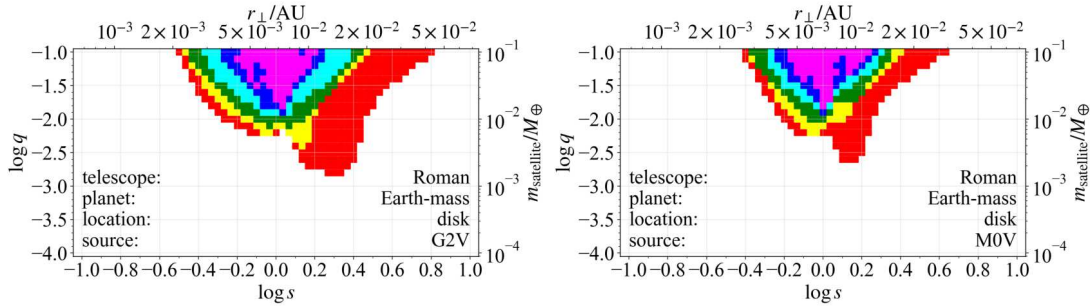
In comparison, the detection efficiency of satellites for an Earth-mass FFP (top-left panel of Figure 4) shows a more asymmetric pattern, with substantially extended sensitivity at  $s > 1$  due to satellite-caustic perturbations broadened by the finite-source effects of the relatively large  $\rho$ . There is modest sensitivity ( $\sim 10\%$  efficiency) to Moon-mass satellites

( $q \approx 0.01$ ) at  $s \approx 1.5$ – $2.5$  ( $r_\perp \sim 0.01$  au). For satellites with about ten times Moon mass ( $q \approx 0.1$ ), the 50% sensitivity zone encompasses  $0.5 \lesssim s \lesssim 2$ , while the 10% zone extends from  $s \sim 0.4$  to  $s \sim 7$ .

The bottom panels of Figure 4 display the results for FFPs in the bulge, showing considerably smaller sensitivity zones than the disk FFPs. This difference arises from two main factors: bulge-lens events have larger  $\rho$ , which tends to reduce the amplitudes of the signals, and events with disk lenses have longer timescales, enabling better light-curve coverage.



**Figure 5.** Satellite Detection efficiency for FFPs (left: Earth; right: Neptune) in the disk with an M0V source.



**Figure 6.** Roman satellite sensitivity for Earth-mass FFP events with G2V (left) and M0V (right) sources.

### 3.2. *M-dwarf Source*

We evaluate CSST satellite detection efficiency using an M0V star as the source star, shown in Figure 5. In comparison to a G2V star, an M0V source is significantly fainter, making the satellite signals more difficult to detect. Additionally, the smaller size of the source further reduces the sensitivity zone at  $s > 1$ . The left panel of Figure 5 illustrates the results for an Earth-mass FFP lens in the disk with an M0V source, revealing limited sensitivity, capable of detecting only satellites with relatively large mass ratios ( $q \sim 0.1$ ) near  $s \sim 1$ . For a Neptune-class lens (right panel of Figure 5), the sensitivity zone in  $\log q$  is still substantial, but it is no longer sensitive to Moon-mass satellite ( $\sim 0.01 M_{\oplus}$ ).

### 3.3. *Roman*

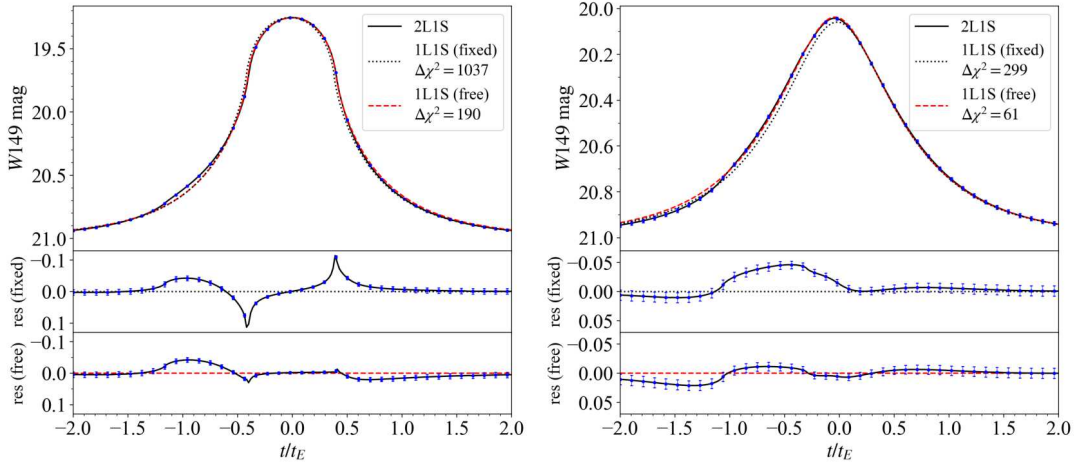
We compute the satellite detection efficiency for Roman. As shown in Figure 6, Roman has greater satellite sensitivity than CSST. For an Earth FFP in the disk with a G2V source (the left panel of Figure 6), Roman's sensitivity extends to Moon-mass satellites near  $s = 1$  and even some sub-Moon-mass (down to  $\sim 0.2 M_{\text{Moon}}$ ) satellites at wide separations ( $s \sim 2$ , approximately 0.02 au). In comparison, with an M0V source, Roman's detection efficiency (the right panel of Figure 6) decreases only modestly, unlike CSST's more drastic drop (the top-left panel of Figure 4 in compared to the left panel of Figure 5), as discussed in Section 3.2. This is primarily because Roman observes in the infrared, which is significantly more sensitive

to the M-dwarf spectral energy distribution than the optical CSST observations.

Sajadian & Sangtarash (2023) investigated the detectability of FFPs' satellites with Roman, focusing on analogs to 25 planet-satellite pairs in the solar system and evaluating systems with  $\log q$  uniformly distributed over the range of  $[-9, -2]$ . Their reported detection efficiency is markedly higher than that derived from our approach. Sajadian & Sangtarash (2023) did not specify how they fitted the mock data with 1L1S models to calculate the  $\Delta\chi^2$  values. We find that fixing the 1L1S parameters ( $t_0$ ,  $u_0$ ,  $t_E$ ) to those of the underlying 2L1S models is needed to reproduce the difference between 2L1S and 1L1S models illustrated in the residual sub-panels of their Figure 2. As shown in Appendix B of Dong et al. (2006), this approach can significantly overestimate the detection efficiency of planets. Therefore, treating the 1L1S parameters as free parameters is necessary to properly simulate the detection process, as implemented in our analysis. Figure 7 provides two examples from Sajadian & Sangtarash (2023), both exhibiting significantly larger  $\Delta\chi^2$  values than those using our procedure of freely fitting 1L1S models. Consequently, we infer that Sajadian & Sangtarash (2023) overestimated their satellite sensitivity.

### 3.4. *Impact Parameter*

The main analysis assumes zero impact parameter  $u_0 = 0$ , and in this section, we examine the effects of a non-zero  $u_0$ .



**Figure 7.** Two examples as shown in (a) and (b) panels of Figure 1 in Sajadian & Sangtarash (2023). Both share the same  $t_E = 0.1$  day,  $q = 0.03$ ,  $s = 0.7$ , and  $\rho = 0.4$ , while they differ in  $u_0$  and  $\alpha$ . We generate the mock data (blue dots) based on the 2L1S models (solid black lines) and fit them to two sets of 1L1S models: fixed (black dotted lines) and free (red dashed lines) parameters. In the fixed-parameter set, the parameters are fixed to be identical to those underlying the 2L1S model, in the same fashion as in Sajadian & Sangtarash (2023). The free-parameter set allows the parameters to vary freely, yielding the best-fit  $\Delta\chi^2$  values (as indicated in the legends) that are significantly smaller than the former set. The residuals to the fixed-parameter and best-fit free-parameter models are shown in the bottom two sub-panels, respectively. Note that Sajadian & Sangtarash (2023) introduced randomized scatters to the data points, preventing exact reproduction of their  $\Delta\chi^2$  values. However, this does not impact our primary conclusion regarding the comparison between fixed and free models.

In Figure 8, we show CSST satellite detection efficiency for an Earth-mass FFP lens in the disk with a G2V source ( $\rho = 0.33$ ) and a Neptune-class disk lens with an M0V source ( $\rho = 0.1$ ), in the first and second rows of panels, respectively. The left, middle, and right sub-panels correspond to  $u_0 = 0, \rho/2, \rho$ , respectively. For cases with  $u_0 > 0$ , we present the efficiency difference relative to  $u_0 = 0$  in the third and fourth rows. Most sensitivity zones show no significant differences, with the grazing case ( $u_0 = \rho$ ) exhibiting a distinct reduction in efficiency ( $\sim 40\%$ ) in only a small portion of the sensitivity zones. Therefore, our default analysis with  $u_0 = 0$  effectively represents the overall cases.

#### 4. Existence of Simulated Satellites

The lack of detected exomoons leaves us without evidence to determine whether planet-satellite systems in the solar system are representative or not. This situation is analogous to the pre-1990s era of exoplanet searches, when the solar system was the only known planetary system. Drawing from the lessons of diverse exoplanet discoveries, we adopt an approach that probes the available parameter space without making a priori assumptions about the distribution of exomoons. Nevertheless, the existence of detectable exomoons is subject to certain physical constraints. In this section, we examine simple physical considerations, including the Roche radius, the Hill radius, and the survival of exomoons during dynamical ejections.

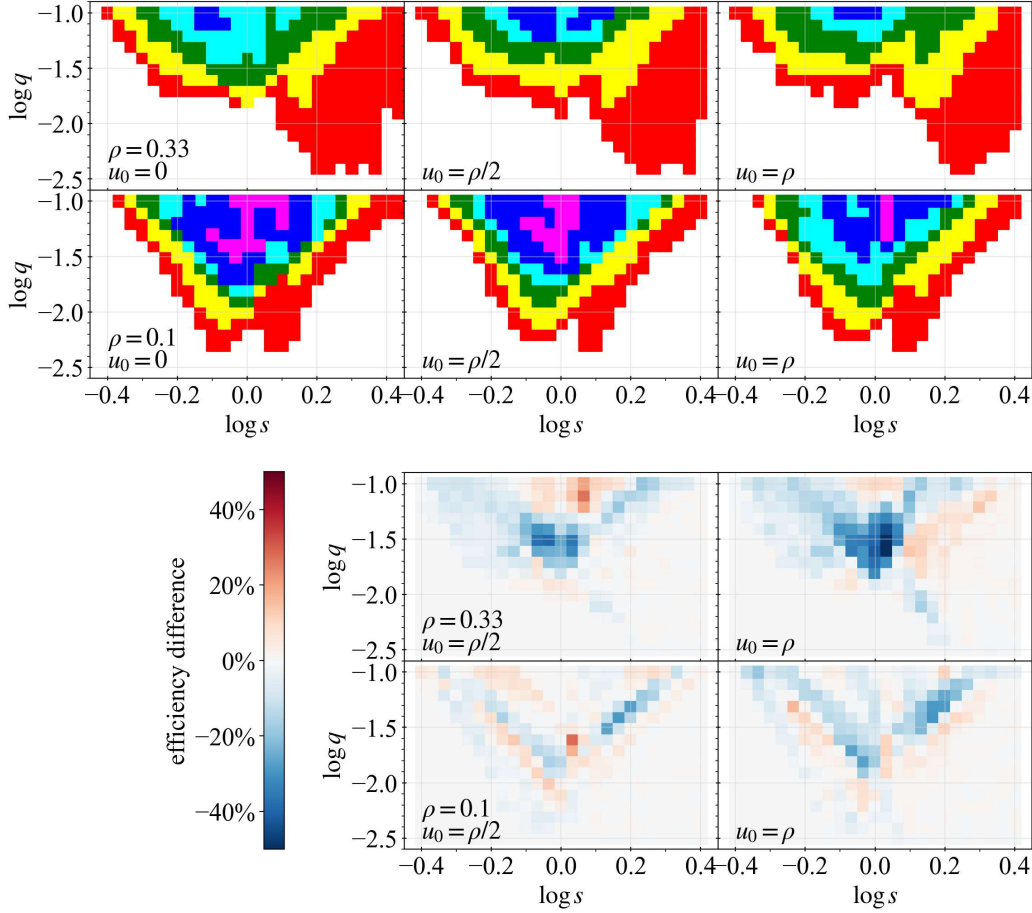
The minimum orbital separation for a satellite to avoid tidal disruption is given by the planet's Roche radius,  $r_{\text{Roche}} = 2.44 R_{\text{planet}} (\rho_{\text{planet}} / \rho_{\text{satellite}})^{1/3}$ , where  $R_{\text{planet}}$  is the planet's radius and  $\rho_{\text{planet}} (\rho_{\text{satellite}})$  is the density of the planet (satellite). For typical densities of planets ( $\sim 3\text{--}6 \text{ g cm}^{-3}$  for rocky planets,  $\sim 0.7\text{--}1.6 \text{ g cm}^{-3}$  for gas or ice giants) and satellites ( $\sim 1.5\text{--}3.5 \text{ g cm}^{-3}$ ),  $r_{\text{Roche}}$  is several times  $R_{\text{planet}}$ , much smaller than the minimum separations detectable planets by CSST or Roman. Therefore, the Roche radius constraint has no impact on our parameter space of interest.

A satellite remains stably bound to a planet within the Hill radius:

$$r_H = a_{\text{planet}} \left( \frac{m_{\text{planet}}}{3m_{\text{star}}} \right)^{1/3} = 0.01 \text{ au} \left( \frac{a_{\text{planet}}}{1 \text{ au}} \right) \left( \frac{m_{\text{planet}}}{M_{\oplus}} \right)^{1/3} \left( \frac{m_{\text{star}}}{M_{\odot}} \right)^{-1/3}, \quad (3)$$

where  $a_{\text{planet}}$  is planet's semimajor axis, and  $m_{\text{star}}$  is the mass of the host star. Dynamical simulations (Debes & Sigurdsson 2007; Hong et al. 2018; Rabago & Steffen 2019) suggest that, the semimajor axes of planet-satellite systems generally experience only minor changes during planet-planet scattering. Thus, we assume that the ejected systems retain their original semimajor axes.

The Einstein radii of our simulated FFPs in the disk (bulge) are  $R_E = D_L \sqrt{\kappa M_L \pi_{\text{rel}}} = 0.007 (0.005) \text{ au} (m_{\text{planet}} / M_{\oplus})^{1/2}$ . The scaled Hill radius for the disk (bulge) cases is then given



**Figure 8.** The first (second) row presents the CSST satellite detection efficiency in  $s$ - $q$  parameter space, with three different impact parameters  $u_0 = 0, \rho/2, \rho$ , for an Earth-mass (Neptune-class) lens in the disk with a G2V source (M0V source). The color coding indicates efficiency levels: 10% (red), 25% (yellow), 50% (green), 75% (cyan), 90% (blue) and 100% (magenta) in the first and second rows. For cases with  $u_0 > 0$ , the third and fourth rows show the efficiency differences relative to the  $u_0 = 0$  cases.

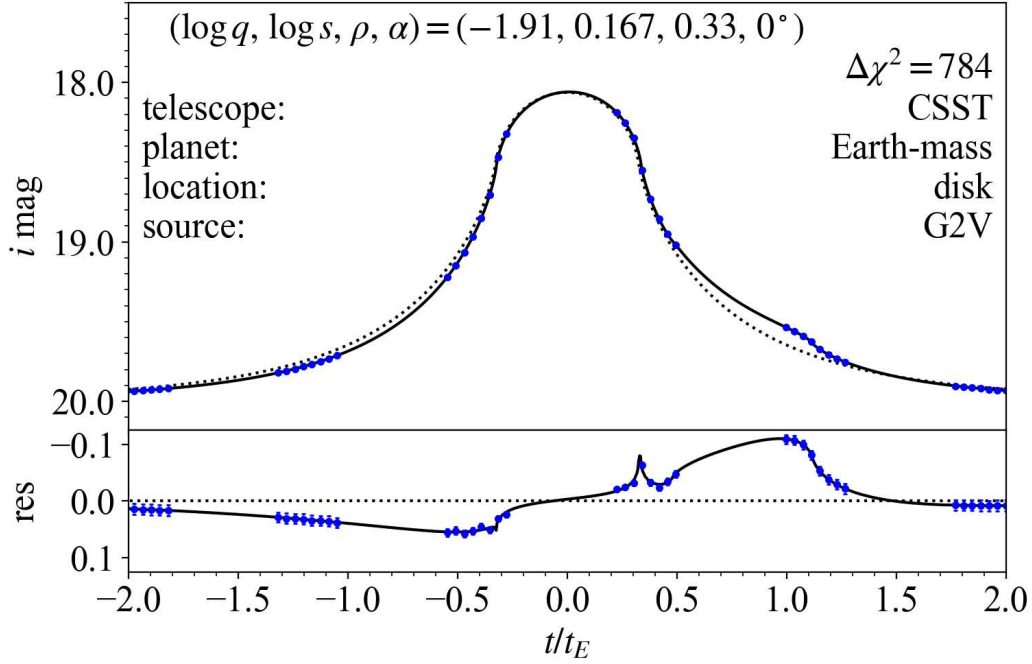
by:

$$\frac{r_H}{R_E} = 1.4 (2.0) \left( \frac{a_{\text{planet}}}{1 \text{ au}} \right) \left( \frac{m_{\text{planet}}}{M_{\oplus}} \right)^{-1/6} \left( \frac{m_{\text{star}}}{M_{\odot}} \right)^{-1/3}. \quad (4)$$

Therefore,  $r_H/R_E = 1$  corresponds to Earth-mass planets orbiting a  $1 M_{\odot}$  star at  $a_{\text{planet}} = 0.7$  (0.5) au in the disk (bulge). The 50% satellite detection zone extends up to at most a factor of  $\sim 2$  (3) from  $R_E$  for CSST (Roman). Thus, for systems ejected from  $a_{\text{planet}} \lesssim 1$  au, only a minority fraction of detectable satellites in the outer part ( $s \gg 1$ ) of the sensitivity zone are outside the Hill radius. Figure 9 illustrates an example of a Moon-mass satellite detectable by CSST within the Hill radius of an Earth analog. For wide-separation ( $\gtrsim 20$  au) Earth-mass planets, the entire sensitivity zone lies within the Hill radius. For Neptune-class FFPs in the disk (bulge),  $r_H/R_E = 1$  corresponds to  $a_{\text{planet}} = 0.013$  (0.008) au, much smaller than typical orbital radius ( $\gtrsim 1$  au) for ejection. Therefore, the satellite detection zones for Neptune-class FFPs are entirely

within the Hill radius. Note that microlensing measures the projected satellite-planet separation on the sky, and consequently, factors like inclination and eccentricity need to be taken into account in a realistic case.

Previous studies (Hong et al. 2018; Rabago & Steffen 2019) have shown that some planet-satellite systems may retain their satellites during dynamical ejection. Hong et al. (2018) used  $N$ -body simulations to investigate the dynamics of exomoons during planet-planet scattering, finding that most satellites with  $a \lesssim 0.1 r_H$  survive post-scattering for planets of  $0.1$ – $1 M_J$ . Similarly, Rabago & Steffen (2019) showed that in systems with three Jupiter-mass planets orbiting a solar-mass star, bound satellites typically have  $a \lesssim 0.07$  au (i.e.,  $\lesssim 0.1 r_H$ ) after scattering. If this survival threshold of  $\lesssim 0.1 r_H$  applies to lower-mass FFPs in our simulations, most of our detectable satellites around Neptune-class planets will survive. In contrast, most satellites in the sensitivity zone for Earth-mass FFPs do not survive if ejections occur in the inner planetary



**Figure 9.** A mock CSST light curve of a detectable Moon-mass satellite orbiting an Earth-mass FFP in the disk. The satellite-planet separation is  $r_{\perp} \approx 0.007$  au, within one (one tenth) Hill radius of an Earth-mass planet at 1 (10) au.

system at  $\sim 1$  au. However, dynamical considerations suggest that ejections are more likely to take place in the outer planetary system, where the Safronov number exceeds unity, and most detectable satellites ejected at  $\gtrsim 10$  au survive.

## 5. Summary and Discussion

In this paper, we study CSST detection efficiency for satellites around Earth-mass and Neptune-class FFPs, considering two representative types of G2V (solar-like) and M0V (M-dwarf) stellar sources. For a G2V source, CSST is capable of detecting satellites around a Neptune-class FFP in the Galactic disk down to Moon-mass satellites near the Einstein radius, with the sensitivity zone extending over a decade in projected separation for an Earth-mass satellite. The detection efficiency in the  $s$ - $q$  plane decreases significantly for an Earth-mass FFP. Its sensitivity zones show a pronounced bi-modal shape, including some sensitivity to Moon-mass satellite at  $s \sim 2$ . By comparison, the satellite detection efficiency reduces substantially for FFPs in the Galactic bulge or for an M0V source. Roman demonstrates higher sensitivity compared to CSST, particularly for M-dwarf sources, due to its infrared bandpass. The exomoon detections enabled by CSST and Roman will probe uncharted territories of exomoons around FFPs and test theoretical predictions of planetary dynamical evolutions.

In our CSST simulations, we assume an idealized observing strategy of continuous  $i$ -band exposures during a 40 minutes span in each orbit. After the microlensing survey strategy is

finalized, more sophisticated simulations will be necessary to incorporate realistic observing patterns and the effects of multiple filters. We also note that some binary-lens perturbations could be mimicked by binary-source single-lens (1L2S) events (e.g., Gaudi 1998). Discussion on the possible 2L1S–1L2S degeneracy is deferred to future studies.

Finally, we briefly discuss the possibility of tidal heating. Ejection of planet-satellite systems generally results in eccentric orbits (Rabago & Steffen 2019), which could lead to significant heating due to tidal circularization. Debes & Sigurdsson (2007) showed that an ejected Earth-Moon system with a semimajor axis  $a \sim 0.001$  au experiences significant tidal heating—exceeding radiogenic heating on the Earth during the first few  $10^8$  yr—potentially making it habitable. However, such close separations are not detectable based on our simulations. Neptune-Earth systems at detectable separations ( $a \sim 0.01$  au) could achieve similar heating rates from tidal circularization, governed by  $|\dot{E}_{\text{circ}}| \propto (R_{\text{planet}}/a)^5 a^{-5/2} m_{\text{satellite}}^2 \sqrt{m_{\text{satellite}} + m_{\text{planet}}}$  (Debes & Sigurdsson 2007). Therefore, spaced-based microlensing surveys are likely promising for detecting tidally heated Earth-mass satellites around free-floating/wide-separation icy giants, if they exist.

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### ORCID iDs

Haozhu Fu  <https://orcid.org/0009-0002-4542-8046>

Subo Dong  <https://orcid.org/0000-0002-1027-0990>

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